

ASEN 5052-001, 5052-001B Analytical Astrodynamics

TTh, 10:05-11:20, AERO 111

**Instructor:** Daniel Scheeres, [scheeres@colorado.edu](mailto:scheeres@colorado.edu)  
AERO 454  
Office Hours TBD

Introduction to astrodynamics with an emphasis on analytical approaches — alternative to ASEN 5050. General solution of the 2-body problem. Orbital trajectories, transfers, targeting, and time of flight. Orbit perturbations and averaging analysis. Restricted 3-body problem.

**Pre-requisite:** Undergraduate orbital mechanics course (equivalent to ASEN 3200) or permission of the instructor.

**Coursepack:**

Selected excerpts from “Orbital Motion in Strongly Perturbed Environments” will be distributed, selected papers will be distributed.

**Textbooks:**

A.E. Roy, Orbital Motion 4th edition, Institute of Physics Publishing, 2005.

**Additional Reference Books:**

D.J. Scheeres. “Orbital Motion in Strongly Perturbed Environments: Applications to Asteroid, Comet and Planetary Satellite Orbiters,” Springer-Praxis Books in Astronautical Engineering. 2012. ISBN 978-3-642-03255-4, e-ISBN 978-3-642-03256-1, DOI 10.1007/978-3-642-03256-1

**Grading:**

|                         |     |
|-------------------------|-----|
| HW problems:            | 25% |
| Computational problems: | 25% |
| Mid-term exam:          | 25% |
| Final exam:             | 25% |

**Topics:**

Principles of orbital mechanics.  
Orbital trajectories, transfers, time of flight.  
Trajectory propagation and targeting.  
Orbit perturbation formulation and analysis.  
Restricted 3-body problem with applications.

**Syllabus (Scheeres):**

## Orbital mechanics

Formulation of two-body, three-body and n-body problems  
The two-body problem solution  
Elliptical and circular orbits  
Parabolic and hyperbolic trajectories  
3-D trajectories and orbit elements  
Time of flight and orbit propagation

## Orbital transfers

Impulsive maneuvers  
Lambert's theorem  
3-D Targeting  
Fuel optimal considerations

## Orbit perturbation formulations

Variation of constants  
Lagrange's Equations  
Gauss' Equations  
Mean elements and averaging

## Orbit perturbation analysis

Effect of non-spherical gravity fields  
Low-thrust trajectories  
Atmospheric drag  
Tidal and third body effects

## Restricted 3-body problem with applications

Derivation of equations of motion  
Jacobi Integral, Zero-Velocity Curves, and Lagrange Points  
Hill approximation  
Numerical computation and analysis of orbits!

**In-Class vs Remote course access:**

The following items detail my plans for delivering lectures and office hours, accommodating any restrictions that may arise from the current pandemic crisis. If the campus transitions to a more restrictive stage, the course has been designed to be able

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: ; G0A4; E: ; 5L''% =4AE<H4; '4; 'A: X8: 7H; F'<>>4EE49<H4; 7'07'4><5: 9'4; '56: '&C7<D0705@'(: AGC>: 7'K: D705: L'  
04; 5<>5'&C7<D0705@'(: AGC>: 7'<5']\_]TbV^TcadU'4A'97C; =4e>4?4A<94L: 98'=4A'=8A56: A'<77C75<; >; L''%@48'6<G: '<  
5: EB4A<A@'E: 9C<?>'4; 9C0H4; J'7: : '-: EB4A<A@' : 9C<?>'04; 9C0H4; 7'4; '56: '&C7<D0705@'(: AGC>: 7'K: D705: L'

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<; 9'BA4; 48; 7'<A: '7C75: 9'4; 'C; 75A8>54A7g'><77'A475: A7L'% '56: '<D7: ; >; '4='78>6'8B9<5: 7J'56: '; <E: '56<5'  
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8; <8564ACf: 9'<>>: 77'54'<<<9: EC>'E<5: AC<?7J'>'C>Y: A'=A<89J'78DECI ; F'56: '7<E: '4A'7CEC?<A'K4AY'C; '  
E4A: '56<; '4; : '>48A7: 'K056485'B: AEC77C4; '=A4E'<??'>48A7: 'C; 75A8>54A7'C; G4?G: 9J'<; 9'<C9C; F'  
<<<9: EC>'9C764; : 75@L'+??'C; >C9: ; 57'4='<<<9: EC>'EC7>4; 98>5'KC??'D: 'A: B4A5: 9'54'56: '24; 4A'049: '  
j64; 4Ae>4?4A<94L: 98kh']\_]TbV^Tl l l \_kl'(589: ; 57'=-48; 9'A: 7B4; 7CD?: '=4A'GC4?<H; F'56: '<<<9: EC>'  
'C; 5: FAC5@'B4?C>@'KC??'D: '78DM: >5'54'; 4; <<<9: EC>'7<; >H4; 7'=A4E'56: '24; 4A'049: '<7'K: ??<7'  
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K: ?>4EC; F'?; <A; C; FJ'K4AYC; FJ'<; 9'7CGC; F': ; G0A4; E: ; 5L'O\$', 48?9: A'KC??'; 45'54?: A<5: '<>57'4='7: 08<?'  
EC7>4; 98>5'j6<A<77E: ; 5J': 0B'4C5<H4; J'<; 9'<77<8'5KJ'C; HE<5: 'B<A5; : A'GC4?: ; >; 'j9<H; F'4A'  
94E: 7H>'GC4?: ; >; kJ'75<'YC; FJ'4A'BA45: >5: 9T>?<77'9C7>ACEC; <H4; '4A'6<A<77E: ; 5'D@'4A'<F<C; 75'  
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8; CG: A705@'B4?C>C: 7J'

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<; 9'=<A?'K56'<?'7589: ; 57'K64J'D: ><87: '4='A: ?FC487'4D?CF<H